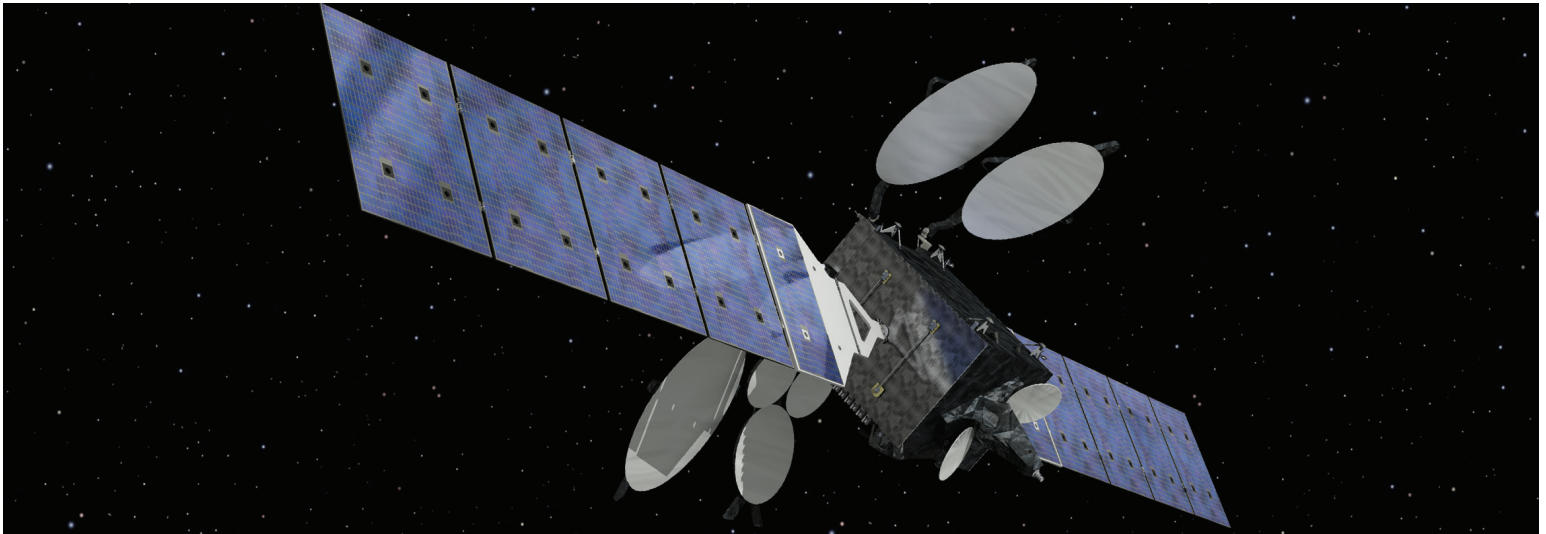


# GEOStar™-3 Bus

A Fully Redundant Spacecraft Bus Designed for Geosynchronous Missions

## FACT SHEET



### Design

The GEOStar-3 satellite platform represents an evolutionary growth of Orbital ATK's GEOStar-2 platform, providing an incremental expansion of the flight-proven GEOStar-2 product line. EPS enhancements include an increase in battery capacity and in solar array power, enabling GEOStar-3 to provide up to 8,000 watts of power to the payload at end-of-life. The larger solar arrays and additional battery capacity retain the 100 percent successful flight heritage 36 volt regulated power bus. The GEOStar-3 Bus structure's mass carrying capability and propellant tank accommodation enable optimal use of launch vehicle performance, and can include tandem launch missions that use heritage bi-propellant apogee engines to ensure fast and reliable orbit raising. For heavier missions, a flight proven Electric Propulsion system will replace the heritage ImpEHTs for stationkeeping operations.

### Payload Accommodations

The GEOStar-3 Bus can accommodate payloads of up to 800 kilograms and 8,000 watts. As many as 72 radiatively cooled TWTAs, or a greater combination of radiatively and conductively cooled TWTAs can easily be accommodated on the platform. The total payload conductive thermal dissipation is scalable up to 4,800 watts. The platform can accommodate a broad range of deployable reflectors and nadir antenna configurations, enabling compelling payload solutions. The expanded payload envelope of GEOStar-3 also allows customers to consider a larger universe of hosted payload opportunities.

### FACTS AT A GLANCE

- Increased power system capability to provide 8 kW of payload power.
- Structure accommodates up to 800 kg payload mass and 4.8 kW payload conducted thermal dissipation.
- Compatibility with all commercially available launch vehicles allowing greater flexibility and lower launch cost.
- Ability to dual manifest two spacecraft as a tandem launch on select launch vehicles.
- Incorporation of flight-proven, low-risk components.
- Configure-to-order flexibility to optimize customer value.

## Specifications

### Core Bus Features

Payload Mass Capability:	Up to 800 kg (single)
Orbit:	Geosynchronous
Typical Mission Lifetime:	>15 years
Delivery:	24 months after receipt of order (payload dependent)
Single Launch	
Compatibility:	Ariane 5, Falcon 9, H-IIA, Proton, Sea Launch, Land Launch
Dual Launch	
Compatibility:	Ariane 5, Proton, Falcon Heavy

### Structure

Bus Dimensions (HxWxL):	3.0 to 3.9 m* x 2.1 m x 2.3 m
Construction:	Composite/Al

### Power Subsystem

Payload Power:	8,000 W
Bus Voltage:	36 VDC (nominal)
Solar Arrays:	Multi-junction GaAs cells
Batteries:	Li-Ion

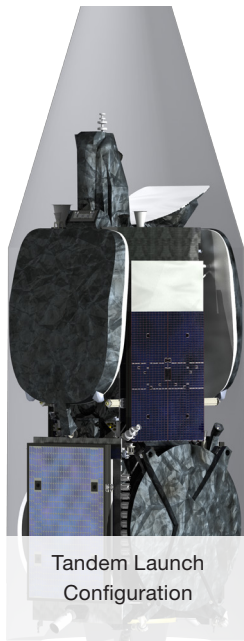
### Attitude Control Subsystem

Stability Mode:	3-axis; zero momentum Propulsion Subsystem
Transfer Orbit System:	Liquid bi-propellant
On Orbit:	Electric + Monopropellant

### Command & Data Handling Subsystem

Flight Processor:	BAE RAD750
Interface Architecture:	MIL-STD 1553B, CCSDS

\*Extensible payload module height



Tandem Launch  
Configuration

## Key Contact

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## Low Cost Launch Opportunities

The GEOStar-3 Bus is designed to maintain compatibility with all commercially available launch vehicles. The ability to launch on multiple vehicles gives customers greater flexibility and lower launch cost access to space. The propellant tanks on GEOStar-3 support a larger delta-V to orbit than typical GEO satellites, which allows GEOStar-3 to make optimal use of launch vehicle performance capability. In addition, GEOStar-3 adds the opportunity to dual-manifest two spacecraft on select launch vehicles, with each spacecraft configured to fulfill its mission independently after launch. This creates an opportunity for dramatic launch cost savings. Whether designed for a single-launch or dual-launch campaign, GEOStar-3 offers a significant increase in payload mass capability.

## Mission Services

Customers can purchase a GEOStar-3 spacecraft bus alone, or as part of a turn-key service that includes an integrated satellite, operations and launch vehicle. Orbital ATK conducts spacecraft commissioning from its own ground station prior to transferring spacecraft control to the customer's operations center.

## Mission Life

As with the GEOStar-2, the GEOStar-3 satellite is designed with conservative margin beyond 15 years, taking into account damaging effects of the geosynchronous environment. The typical limitation of mission duration is on-board fuel for orbit maintenance station-keeping. Fuel life can be optimized and extended to well over 15 years, based on launch vehicle selection. Orbital ATK has several delivered satellites with predicted fuel life in excess of 20 years. GEOStar-3 has propellant accommodation that can be used to significantly extend maneuver life.

## Heritage and Versatility

GEOStar-3 was designed around the concept of scalability and customer options, while still maintaining the use of standardized, qualified components and subsystems. The C&DH system of the GEOStar family has been upgraded to provide more capability to support advanced mission requirements. The EPS system of the GEOStar family, while retaining the heritage 36V regulated bus architecture, offers expandable capability to support GEOStar-3 class payload power requirements. The dual-manifest launch capability introduced with GEOStar-3 uses the same highly reliable, short duration bi-propellant orbit raise propulsion system that has enjoyed success on over 30 GEOStar-2 spacecraft. GEOStar-3 has options for battery and solar array size, payload module height and station-keeping thruster selection that allow Orbital ATK to provide customers with the best balance of satellite and launch cost, mission lifetime and payload performance. With its expanded payload carrying capability, enhanced C&DH functionality and larger EPS system, GEOStar-3 also offers customers the largest hosted payload envelope in the GEOStar family.

## Value

GEOStar-3 further enhances Orbital ATK's value proposition to its customers, combining fast delivery, low cost and high reliability with a larger payload envelope and low cost launch alternatives.